

ENERGY REQ'D TO BRING UNIT MASS INTO SATELLITE ORBIT

$$E = \underbrace{\frac{1}{2} V_s^2}_{(1)} + \underbrace{\int_{R_0}^R g \, dR}_{(2)}$$

TERM (1) - see eqn. 5-30 $= \frac{1}{2} R_0^2 \frac{g_0}{R_0+h}$ ✓

TERM (2)

$$\int_{R_0}^R g \, dR = \int_{R_0}^R g_0 \frac{R_0^2}{R^2} \, dR \quad (\text{see eqn. 5-7})$$

$$= g_0 R_0^2 \int_{R_0}^R \frac{1}{R^2} \, dR = g_0 R_0^2 \left[-\frac{1}{R} \right]_{R_0}^R$$

$$= g_0 R_0^2 \left(-\frac{1}{R} + \frac{1}{R_0} \right)$$

since $R = R_0 + h$

$$= g_0 R_0^2 \left(-\frac{1}{R_0+h} + \frac{1}{R_0} \right)$$

$$= g_0 R_0 - \frac{g_0 R_0^2}{R_0+h}$$

combine terms (1) and (2)

$$E = \frac{1}{2} R_0^2 \frac{g_0}{R_0+h} + g_0 R_0 - \frac{g_0 R_0^2}{R_0+h}$$

$$= -\frac{1}{2} R_0^2 \frac{g_0}{R_0+h} + g_0 R_0$$

$$= \frac{-\frac{1}{2} R_0^2 g_0 + g_0 R_0 (R_0+h)}{R_0+h}$$

$$= \frac{-\frac{1}{2} R_0^2 g_0 + \frac{1}{2} g_0 R_0 (2R_0+2h)}{R_0+h}$$

$$= \frac{1}{2} R_0 g_0 \frac{(-R_0 + 2R_0 + 2h)}{R_0+h}$$

$$E = \frac{1}{2} R_0 g_0 \frac{R_0+2h}{R_0+h}$$

EQN. (5-32)

(2)